5. The engine of claim **1** wherein:
a hub-to-tip ratio (R_I:R_O) of the low pressure turbine section is between about 0.4 and about 0.5 measured at the maximum R_O axial location in the low pressure turbine section.
6. The engine of claim **5** wherein:
the low pressure turbine section has 2 to 3 blade stages.
7. The engine of claim **5** wherein:
a ratio of maximum gaspath radius along the low pressure turbine section to maximum radius of the fan is less than about 0.50.
8. The engine of claim **5** wherein:
an airfoil count of the low pressure turbine section is below about 1600.
9. The engine of claim **1** wherein:
a ratio of maximum gaspath radius along the low pressure turbine section to maximum radius of the fan is less than about 0.55.
10. The engine of claim **9** wherein:
said ratio of maximum gaspath radius along the low pressure turbine section to maximum radius of the fan is less than about 0.50.
11. The engine of claim **10** wherein:
said ratio of maximum gaspth radius along the low pressure turbine section to maximum radius of the fan is between about 0.35 and about 0.50.
12. The engine of claim **1** wherein:
said ratio of low pressure turbine section airfoil count to bypass area ratio is between about 10 and about 150.
13. The engine of claim **1** wherein:
the compressor comprises:
a low pressure compressor section; and
a high pressure compressor section.
14. The engine of claim **13** wherein:
the turbine has a high pressure turbine section coupled to drive the high pressure compressor section.
15. The engine of claim **14** wherein:
there are no additional compressor or turbine sections.
16. The engine of claim **13** wherein:
blades of the low pressure compressor section and low pressure turbine section share a shaft; and
the speed reduction mechanism comprises an epicyclic transmission that couples the shaft to a fan shaft to drive the fan with a speed reduction.
17. The engine of claim **1** wherein:
the speed reduction mechanism comprises an epicyclic transmission.
18. The engine of claim **1** wherein:
the low pressure turbine section has 2 to 6 blade stages.
19. The engine of claim **1** wherein:
the low pressure turbine section has 2 to 3 blade stages.
20. The engine of claim **1** wherein:
an airfoil count of the low pressure turbine section is below about 1600.
21. The engine of claim **1** in combination with a mounting arrangement wherein an aft mount reacts at least a thrust load.

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